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## IN THE UNITED STATES PATENT AND TRADEMARK OFFICE

In Re Patent Application of:  
**JONES et al.**

Serial No.: 10/697,945  
Filed: 10/31/03

Group: 3643  
Examiner: COLLINS, Timothy D.

Title: AEROENGINE NACELLE

### PRIORITY CLAIM SUBMISSION AND CERTIFIED COPY

Date: August 22, 2005

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P.O. Box 1450  
Alexandria, VA 22313-1450

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Application No.

0315800.3  
0312490.6

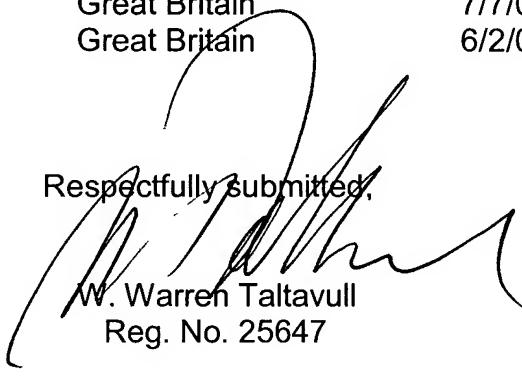
Country of Origin

Great Britain  
Great Britain

Filed

7/7/03  
6/2/03

Respectfully submitted,

  
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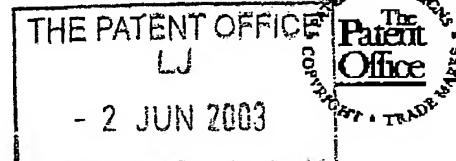
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Rule 16)02JUN03 E811490-1 D00370  
P01/7700 0.00-0312490.6

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1. Your reference  
DY3157

0312490.6

- 2 JUN 2003

2. Patent application number  
(The Patent Office will fill in this part)

ROLLS-ROYCE PLC  
65 BUCKINGHAM GATE  
LONDON  
SW1E 6AT

397 0002

Patents ADP number (if you know it)

If the applicant is a corporate body, give the  
country/state of its incorporation ENGLAND

4. Title of the invention  
AEROENGINE NACELLE

5. Name of your agent (if you have one) M A GUNN

"Address for service" in the United Kingdom  
to which all correspondence should be sent  
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ROLLS-ROYCE plc  
PATENTS DEPARTMENT  
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DERBY  
DE24 8BJ

397 0004

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## Description

8

## Claim(s)

1

## Abstract

1

## Drawing(s)

2

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## Priority documents

## Translations of priority documents

Statement of inventorship and right to grant of a patent (Patents Form 7/77)

2

Request for preliminary examination and search (Patents Form 9/77)

1

Request for substantive examination (Patents Form 10/77)

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Any other documents  
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11.

I/We request the grant of a patent on the basis of this application.

Signature *M A Gunn*

Date

M A GUNN

2.6.2003

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DUPLICATE

ROLLS-ROYCE plc

CASE NO: DY3157

RR TITLE: [REDACTED]

APPLICATION NO:

DATED: 2 JUNE 2003

FIRST APPLICATION

DRAFTED BY: KAW

PATENTS ACT 1977SPECIFICATIONAEROENGINE NACELLE

AEROENGINE NACELLE

The present invention relates to a nacelle for a gas turbine engine and in particular a nacelle configuration comprising an openable thrust reverser unit to provide improved access to accessories mounted on a fan casing and a core engine.

A conventional nacelle for a turbofan gas turbine engine comprises three major components: an intake, fan cowl doors and a thrust reverser unit. These turbofan engines also incorporate a fan containment system attached to an engine fan casing. The fan cowl doors are generally C-shaped that are each opened by a power opening system and hold-open rods for access to fan casing mounted accessories. The thrust reverser unit also comprises two generally C-shaped doors that are each opened by a further power opening system and hold-open rods for access to the core engine and other core mounted engine accessories.

This conventional arrangement is disadvantaged in that there are two independent power opening system and hold-open rods for the fan cowl doors and thrust reverser doors. Furthermore, the prior art nacelle comprises a joint between each section which leads to an undesirable disturbance in airflow over the nacelle inner and outer surfaces with resultant aerodynamic drag losses. This arrangement also requires substantial fire insulation material around the large fan casing chamber inward of the fan cowl doors. Thus the conventional arrangement is heavy and costly to manufacture and is disadvantaged by requiring two sets of doors to be opened for access to the engine.

Therefore it is an object of the present invention to provide a nacelle arrangement that reduces the number of openable doors, reduces weight of the assembly and reduces aerodynamic drag. These are accomplished by reducing the number of nacelle components to two, with primarily the fan cowl doors being integrated partially into the intake and

substantially into the thrust reverser unit. This arrangement also integrates the fan containment system into the nacelle intake removing the need for a bolted flange joint and reducing the size of the fan casing chamber fire zone. As the fan case accessory fire zone is reduced in size less fire extinguishing equipment, ventilation requirements and fire protection treatment is required.

The reduced number of steps or gaps between nacelle components, especially in the external nacelle surface reduces aerodynamic drag. Access to the fan case mounted accessories and core engine is simplified by provision of a single power opening system and hold open rods to the extended thrust reverser unit. The intake of the present invention extends further rearwards and provides an extended length for natural airflow and subsequent aerodynamic drag reduction. By integrating the fan containment casing into the intake, there is no bolted flanged joint forward of the fan system enabling provision for improved acoustic panels thus reducing engine-generated noise.

Accordingly the present invention seeks to provide a nacelle for a gas turbine engine, the engine comprising accessories mounted to a fan casing and a core engine, the nacelle substantially surrounds the engine and comprises an intake and a thrust reverser unit, the thrust reverser unit is formed by two generally C-shaped and openable portions characterised in that the thrust reverser unit is openable to provide access to the accessories and the core engine.

Preferably, the nacelle comprises a fan containment casing, the fan containment casing is integral with the intake.

Preferably, an acoustic panel extends between a downstream edge of the fan containment casing and a leading edge of the intake, thus reducing engine-generated noise.

Preferably, the fan containment casing is disposed outside a fire zone of the nacelle.

The present invention will be more fully described by way of example with reference to the accompanying drawings in which:

Figure 1 is a schematic section of part of a prior art 5 nacelle surrounding a conventional fan gas turbine engine;

Figure 2 is a schematic section of part of a nacelle, in accordance with the present invention, surrounding a conventional fan gas turbine engine;

Figure 3 is an isometric view of a nacelle mounted to 10 a wing and in accordance with the present invention.

With reference to Figures 1 and 2, a ducted fan gas turbine engine generally indicated at 10 has a principal and rotational axis 11. The engine 10 comprises, in axial flow series, an air intake 12, a propulsive fan 13, an 15 intermediate pressure compressor 14, a high-pressure compressor 15, combustion equipment 16, a high-pressure turbine 17, and intermediate pressure turbine 18, a low-pressure turbine 19 and a core nozzle 20. A core duct 22 is partly defined radially inwardly by a core plug 23 and 20 radially outwardly by the core nozzle 20.

The gas turbine engine 10 works in the conventional manner so that air entering the intake 11 is accelerated by the fan 13 to produce two air flows: a first air flow into the intermediate pressure compressor 14 and a second air 25 flow which passes through a bypass duct 24 to provide propulsive thrust. The intermediate pressure compressor 14 compresses the air flow directed into it before delivering that air to the high pressure compressor 15 where further compression takes place.

30 The compressed air exhausted from the high-pressure compressor 15 is directed into the combustion equipment 16 where it is mixed with fuel and the mixture combusted. The resultant hot combustion products then expand through, and thereby drive the high, intermediate and low-pressure 35 turbines 17, 18, 19 before being exhausted through the nozzle 20 to provide additional propulsive thrust. The

high, intermediate and low-pressure turbines 17, 18, 19 respectively drive the high and intermediate pressure compressors 15, 14 and the fan 13 by suitable interconnecting shafts 25, 26, 27.

5 A fan casing 28 circumferentially surrounds the fan 13 and an array of outlet guide vanes 29 and comprises a fan containment casing 33. The fan containment casing 33 surrounds the fan 13 and is attached to the fan casing 28 at a bolted flange joint 44'. The fan casing 28 is 10 supported by the annular array of outlet guide vanes 29. The fan casing 28 is used as a mount for some engine accessories 34 as known in the art.

Referring to Figure 1, a conventional nacelle 21 generally surrounds the engine 10 and comprises the intake 12, fan cowl doors 30, a thrust reverser unit 31 and a bypass exhaust nozzle 32. It is necessary to access the accessories 34 for maintenance and so fan cowl doors 30 are provided. Two generally C-shaped fan cowl doors 30 are rotatably mounted to a pylon suspended from a wing of an 20 associated aircraft and each are openable via a power opening system and hold open rods as known in the art. The fan cowl doors 30 are located via V-shaped groove joints 35, 36 to the intake 12 and thrust reverser unit 31 respectively. The fan cowl doors 30 and the fan casing 28 25 define a chamber 37 in which the engine accessories 34 are housed. The chamber 37 is a fire zone and must therefore be adequately sealed against fire escaping therefrom. The chamber 37 is provided with fire protection material that lines the chamber 37, fire extinguishing equipment and 30 suitable ventilation requirements. Such fire precautions are well known in the art and an example of which is the Rolls-Royce Trent 700 Aeroengine.

Rearward of the fan cowl doors 30, and located via radially outer and inner V-shaped groove joints 36 and 36' 35 to the downstream end of the fan casing 28, is the thrust reverser unit 31. The thrust reverser unit 31 comprises

radially inner and outer walls 38, 39 and housed therebetween a cascade structure 40 and a drive mechanism 41 for operation of the thrust reverser unit in thrust reversing mode. The drive mechanism 41 is mounted to the 5 axially short member 52, which is attached to the fan casing at joint 36' and to fan cowl doors at joint 36. The outer wall 39 joins the axially short member 52, 52' at joint 46 and the inner wall at joint 46'.

The thrust reverser unit 31 further comprises bypass 10 duct blocker doors 42, rotatably mounted at their upstream end to the inner wall 38 and at their downstream end to a rotatably mounted link 43. The operation of the thrust reverser unit 31 is well known and briefly comprises the drive mechanism 41 driving the radially inner and outer 15 walls 38, 39 rearward to expose the cascade structure 40 to the airflow through the bypass duct 24, which is blocked by the blocker doors 42 having been swung radially inwardly by the rearward movement and the rotatable links 43.

It is necessary to access the core 9 of the engine 10 20 to maintain other engine accessories 48 mounted on a core engine casing 49, the compressors 14, 15, turbines 16, 17, 18 and combustor equipment 16 and so the thrust reverser comprises two generally C-shaped portions that are 25 rotatably mounted to a pylon suspended from a wing of an aircraft and are each openable via a power opening system and hold open rods as known in the art.

However, this conventional arrangement is 30 disadvantaged in that there are two independent power opening systems and sets of hold-open rods for the fan cowl doors 30 and thrust reverser unit doors 31. Furthermore, the prior art nacelle 21 comprises joints 35, 35' and 36, 36' between each nacelle section (12, 30, 31) and further joints 46, 46' that cause undesirable breakdown in airflow over the nacelle inner and outer surfaces and subsequent 35 aerodynamic drag losses. This arrangement also requires substantial fire insulation material within the large fan

casing chamber 37 inward of the fan cowl doors 30. Thus this conventional arrangement is also heavy and costly to manufacture and does not provide simple access for engine maintenance.

5 Referring now to Figures 2 and 3 where the engine 10 and its operation is substantially the same as described hereinbefore, a nacelle 55 in accordance with the present invention generally surrounds the engine 10 and comprises an intake 12 and a thrust reverser unit 31. The fan cowl 10 doors 30 of the prior art nacelle 21 have been removed and are replaced mainly by the radially outer wall 39 being extended forwards, thereby surrounding the fan casing 28, to meet with the rearward part of the intake 12 at joint 44, 44'. The intake 12 comprises the fan containment 15 casing 33 in an integral assembly. Therefore as shown in Figure 2 the tip 57 of each fan blade of the fan 13 is substantially parallel with the engine centre-line 11 to permit the intake 12 to be assembled to the fan casing 28.

The thrust reverser 31 of the present invention 20 comprises two generally C-shaped portions 31a, 31b that are rotatably mounted to a pylon 50 suspended from a wing 51 of an associated aircraft, and are each openable via a power opening system 53 and hold open rods 54 as known in the art. However briefly, the power opening system 53 is 25 rotatably mounted to the fan casing 28 and provides a force to open each C-shaped portions 31a, 31b, while the hold open rods 54 are manually fitted to provide a rigid support. Access to the engine accessories 34 and core engine 9, which includes the core accessories 48, is 30 therefore made simpler through this single set of thrust reverser C-shaped doors 31a, 31b.

It should be appreciated that the nacelle 55 arrangement of the present invention reduces the number of nacelle 55 components to two, with primarily the prior art 35 fan cowl doors 30 being integrated partially into the intake 12 and mainly into the thrust reverser unit 31.

This arrangement then integrates the fan containment system 33 into the intake 12 removing the joints 35, 35' and 36. Joint 36' remains in place between the radially inner wall 38 of the thrust reverser unit 31 and the rearward end of 5 the fan casing 28. The joint 44, 44' is positioned rearward of the fan 13, which reduces the size of the fan casing chamber 37 fire zone. As this fan case accessory fire zone is reduced in size a reduced amount of fire extinguishing equipment, ventilation requirements and fire 10 protection treatment is required making the nacelle assembly 55 lighter and cheaper than the prior art nacelle 21. Furthermore, reducing the number of joints reduces the number of steps or gaps between nacelle components, especially in the external nacelle surface which is 15 beneficial in reducing aerodynamic drag and which allows an improved airflow to form over the radially inner and outer surfaces of the nacelle 55. Access to the fan case mounted accessories 34 and core engine 9 is simplified by provision of a single power opening system 53 and hold open rods 54 20 to open each C-shaped portion 31a, 31b of the thrust reverser unit 31. The intake 12 of the present invention extends further rearwards and provides an extended length for a smooth airflow to form and subsequent aerodynamic drag reduction. By integrating the fan containment casing 25 33 into the intake 12, there is no bolted flanged joint 35, 35' forward of the fan system 13. This enables substantially all of a radially inner surface 45 of the intake 12 to comprise acoustic panels 56 which extend between the downstream edge of the fan containment casing 30 33 and the leading edge of the intake 12, thus reducing engine-generated noise.

It should be appreciated by one skilled in the art that other forms of thrust reverser unit are interchangeable with the cascade structure/blocker door 35 style arrangement described herein, whilst remaining within the scope of the present invention. Other thrust reverser

units include those described in and incorporated herein by reference to "The Jet Engine", Rolls-Royce Plc, 1986 pp159-167, and those others as known in the art, such as pivoting door type thrust reversers.

5 Whilst endeavouring in the foregoing specification to draw attention to those features of the invention believed to be of particular importance it should be understood that the Applicant claims protection in respect of any patentable feature or combination of features hereinbefore 10 referred to and/or shown in the drawings whether or not particular emphasis has been placed thereon.

**Claims:**

1. A nacelle (55) for a gas turbine engine (10), the engine (10) comprising accessories (34) mounted to a fan casing (28) and a core engine (9), the nacelle (55) substantially surrounds the engine (10) and comprises an intake (12) and a thrust reverser unit (31), the thrust reverser unit (31) is formed by two generally C-shaped and openable portions 31a, 31b, characterised in that the thrust reverser unit (31) is openable to provide access to the accessories (34) and the core engine (9).
2. A nacelle (55) as claimed in claim 1 wherein the nacelle (55) comprises a fan containment casing (33), the fan containment casing (33) is integral with the intake (12).
3. A nacelle (55) as claimed in claim 2 wherein the intake (12) comprises an acoustic panel (45) that extend between a downstream edge of the fan containment casing (33) and a leading edge of the intake (12), thus reducing engine-generated noise.
4. A nacelle (55) as claimed in any one of claims 2-3 wherein the fan containment casing (33) is disposed outside a fire zone (37) of the nacelle (55).
5. A nacelle (55) substantially as described in this specification and with reference to and as shown in figures 2 or 3 of the accompanying drawings.

ABSTRACTAEROENGINE NACELLE

5 A nacelle (55) for a gas turbine engine (10), the engine (10) comprising accessories (34) mounted to a fan casing (28) and a core engine (9), the nacelle (55) substantially surrounds the engine (10) and comprises an intake (12) and a thrust reverser unit (31). The thrust reverser unit (31) 10 is formed by two generally C-shaped portions 31a, 31b. The thrust reverser unit (31) is openable to provide access to the accessories (34) and the core engine (9). The nacelle (55) further comprises a fan containment casing (33) that is integral with the intake (12).

15

(Figure 2)

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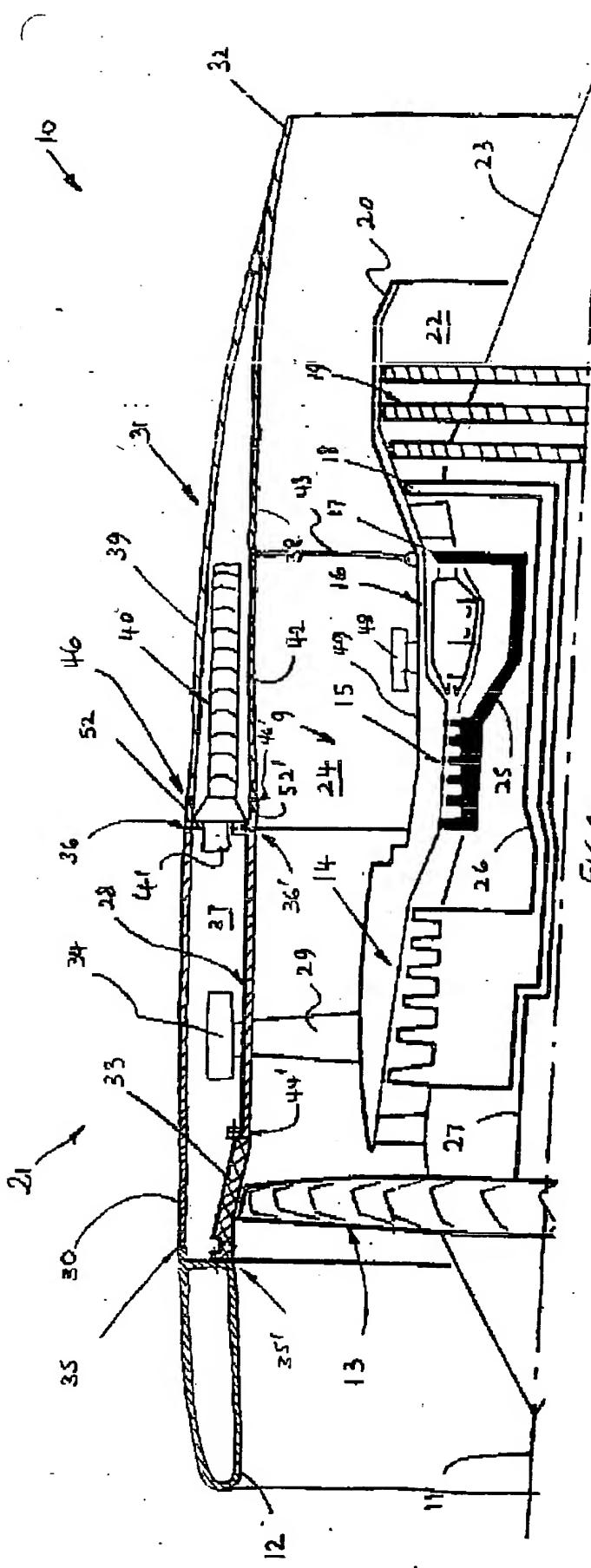


Fig 1

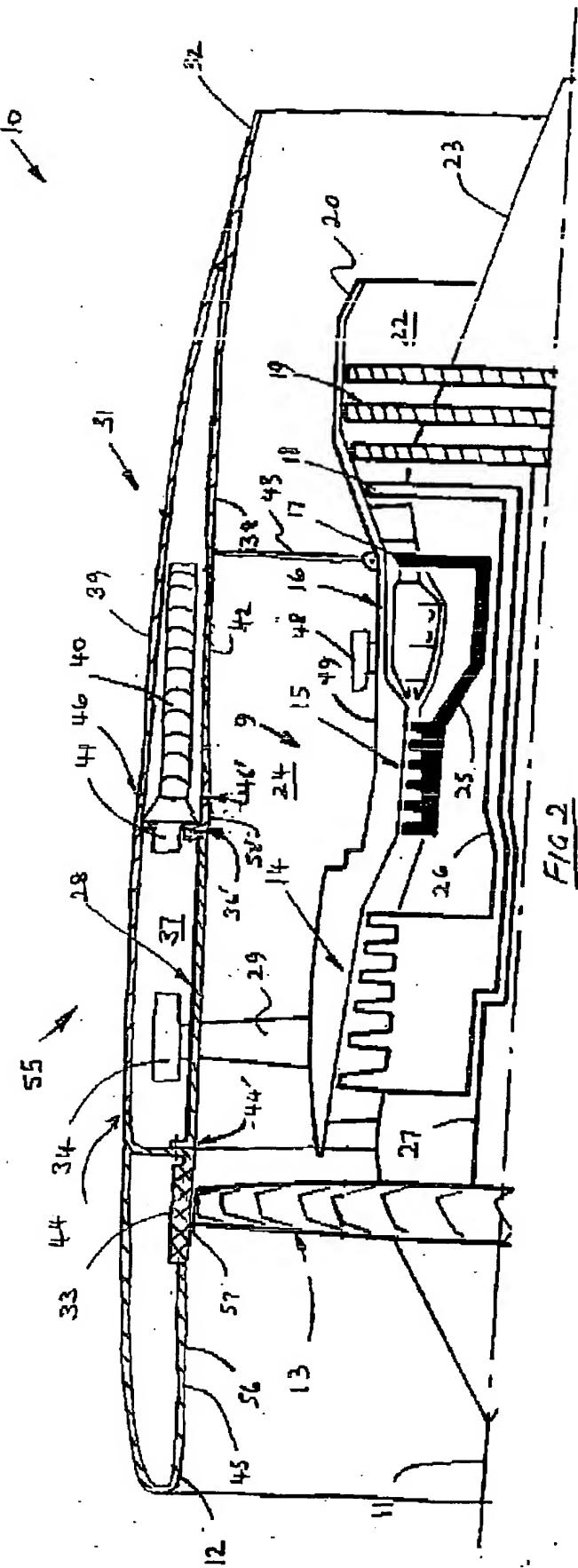


FIG 2

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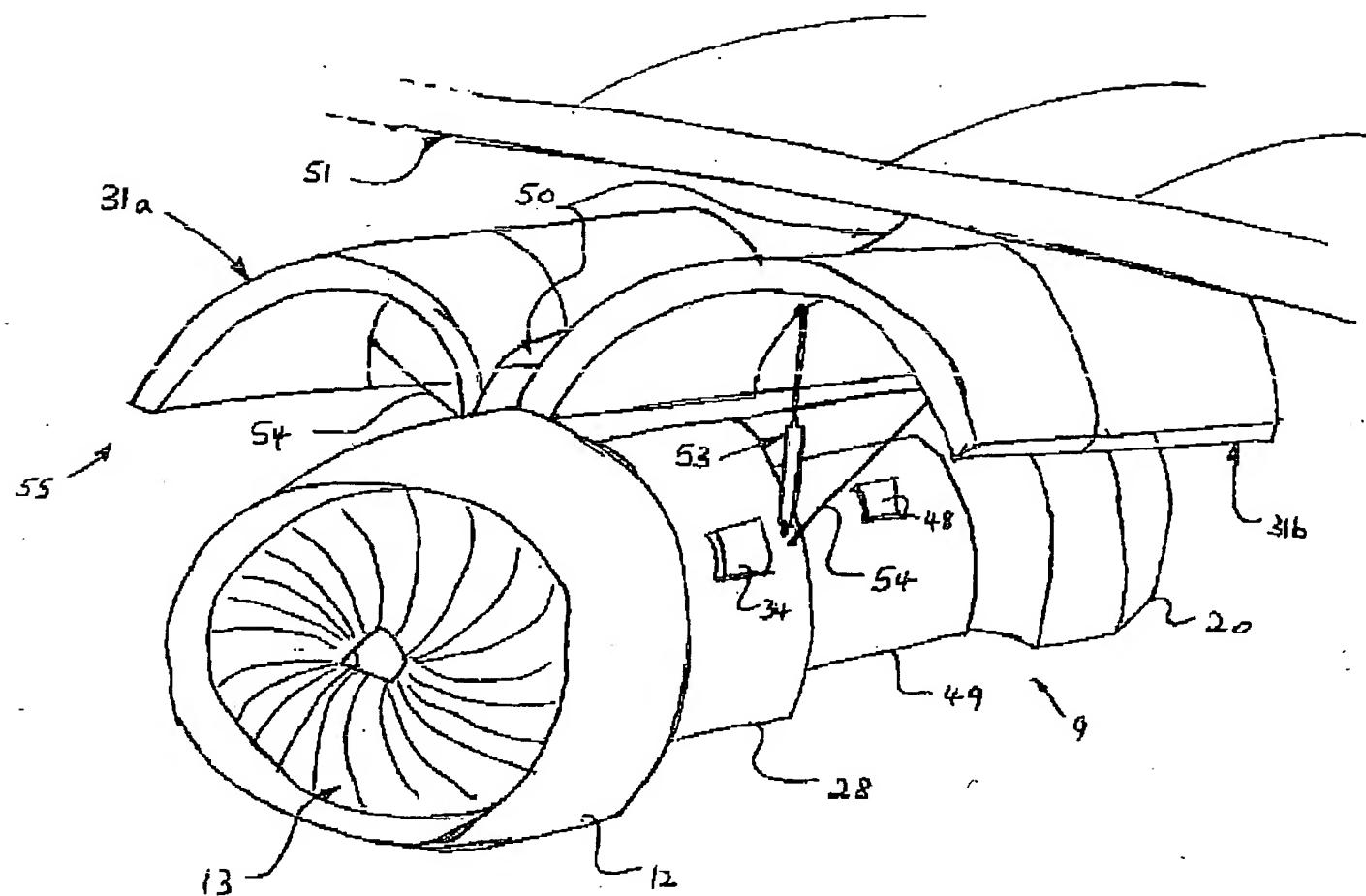


Fig 3

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